

wherein the gas turbine engine is a high bypass geared aircraft engine having a bypass ratio of greater than about six (6).

14. The gas turbine engine according to claim **13**, further comprising:

- a combustor fluidly connected to the compressor section;
- a turbine section fluidly connected to the combustor, the turbine section comprising:
 - a high pressure turbine; and
 - a low pressure turbine.

15. The gas turbine engine according to claim **13**, wherein the core housing includes a first inlet case portion defining an inlet case flow path, and a bearing support portion removably secured to the inlet case portion, the second bearing mounted to the bearing support portion.

16. The gas turbine engine according to claim **13**, wherein the core housing includes an intermediate case portion defining an intermediate case flow path, and a bearing support portion removably secured to the intermediate case portion, the first bearing mounted to the bearing support portion.

17. The gas turbine engine according to claim **13**, wherein the multiple compressor stages includes a variable stator vane array, rotatable compressor blades, and a fixed stator vane array.

18. The gas turbine engine according to claim **13**, wherein said low pressure turbine is one of three turbine rotors, and said low pressure turbine driving said fan, while the other two of said turbine rotors each driving a compressor section

19. The gas turbine engine according to claim **13**, wherein also including a high pressure turbine, with each of said low pressure turbine and said high pressure turbine driving a compressor rotor.

20. The gas turbine engine according to claim **19**, wherein said gear train is positioned intermediate a compressor rotor driven by said low pressure turbine and said fan.

21. The gas turbine engine according to claim **19**, wherein said gear train is positioned intermediate said low pressure turbine and said compressor rotor driven by said low pressure turbine.

22. A gas turbine engine comprising:

- a core housing providing a core flow path;
- a fan;

- a shaft supporting a compressor section arranged within the core flow path, wherein the compressor section is fluidly connected to the fan, the compressor section comprising a first pressure compressor and a second pressure compressor upstream from the first pressure compressor, the second pressure compressor including multiple compressor stages; and

- first and second bearings supporting the shaft relative to the core housing and being arranged radially inward of and axially overlapping with at least some of the multiple compressor stages;

- a combustor fluidly connected to the compressor section;
- a turbine section fluidly connected to the combustor, the turbine section comprising:

- a high pressure turbine;
- a low pressure turbine; and

- wherein the gas turbine engine includes at least one of a low Fan Pressure Ratio of less than about 1.45 and a low pressure turbine pressure ratio that is greater than about 5.

23. The gas turbine engine according to claim **22**, wherein said low pressure turbine is one of three turbine rotors, and said low pressure turbine driving said fan, while the other two of said turbine rotors each driving a compressor section

24. The gas turbine engine according to claim **22**, wherein also including a high pressure turbine, with each of said low pressure turbine and said high pressure turbine driving a compressor rotor.

25. The gas turbine engine according to claim **24**, wherein said gear train is positioned intermediate a compressor rotor driven by said low pressure turbine and said fan.

26. The gas turbine engine according to claim **24**, wherein said gear train is positioned intermediate said low pressure turbine and said compressor rotor driven by said low pressure turbine.

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